

IN THE CLAIMS

Please amend the claims as follows:

Claim 1 (Canceled).

Claim 2 (Currently Amended) ~~The gas turbine according to claim 1,~~ A gas turbine operating at a pressure ratio of 20 or more, comprising a final stage of blades that includes a stationary blade and a moving blade,

wherein said moving blade is constructed such that the pressure difference at the downstream and upstream sides of said moving blade is 0.15 MPa or less, and

a boss ratio at a gas exit side of said moving blade is 0.4 or more to 0.65 or less, wherein the boss ratio is a ratio of a hub radius and a tip radius.

Claim 3 (Canceled).

Claim 4 (Currently Amended) ~~The gas turbine according to claim 1,~~ A gas turbine operating at a pressure ratio of 20 or more, comprising a final stage of blades that includes a stationary blade and a moving blade,

wherein said moving blade is constructed such that the pressure difference at the downstream and upstream sides of said moving blade is 0.15 MPa or less, and

a curvature from a back side throat to a rear edge of said moving blade is 0 or more to 0.15 or less, ~~wherein~~ the curvature is being equivalent to a radius of curvature of the back side from the pitch/throat to the rear edge.

Claim 5 (Original) A gas turbine operating at a pressure ratio of 20 or more, comprising a final stage of blades that includes a stationary blade and a moving blade,

wherein a gauging ratio of said stationary blade is 0.9 or less, wherein the gauging ratio is a ratio of a tip side gauging and a hub side gauging.

Claim 6 (Original) The gas turbine according to claim 5, wherein a boss ratio at a gas exit side of said moving blade is 0.4 or more to 0.65 or less, wherein the boss ratio is a ratio of a hub radius and a tip radius.

Claim 7 (Original) The gas turbine according to claim 5, wherein an average degree of reaction of said final stage of blades is 0.3 or more to 0.6 or less.

Claim 8 (Original) The gas turbine according to claim 5, wherein a curvature from a back side throat to a rear edge of said moving blade is 0 or more to 0.15 or less, wherein the curvature is equivalent to a radius of curvature of the back side from the pitch/throat to the rear edge.

Claim 9 (Original) A gas turbine operating at a pressure ratio of 20 or more, comprising a final stage of blades that includes a stationary blade and a moving blade, wherein an exit angle ratio of said stationary blade is 0.85 or more, wherein the exit angle ratio is a ratio of a tip side exit angle and a hub side exit angle.

Claim 10 (Original) The gas turbine according to claim 9, wherein a boss ratio at a gas exit side of said moving blade is 0.4 or more to 0.65 or less, wherein the boss ratio is a ratio of a hub radius and a tip radius.

Claim 11 (Original) The gas turbine according to claim 9, wherein an average degree of reaction of said final stage of blades is 0.3 or more to 0.6 or less.

Claim 12 (Original) The gas turbine according to claim 9, wherein a curvature from a back side throat to a rear edge of said moving blade is 0 or more to 0.15 or less, wherein the curvature is equivalent to a radius of curvature of the back side from the pitch/throat to the rear edge.

Claim 13 (Original) A gas turbine operating at a pressure ratio of 20 or more, wherein a duct wall in a portion of a specified distance from an end opposing said gas turbine, of a duct forming a diffuser passage communicating with a final exit side of said gas turbine is drawn parallel to or inside of a shaft of the gas turbine.